

**SSME FMEA/CIL
REDUNDANCY SCREEN**

Component Group: Orifices
 CIL Item: N714-01
 Part Number: RS009100
 Component: FPB Oxidizer ASI Downstream Orifice (O16.1.3)
 FMEA Item: N711, N714
 Failure Mode: Orifice restricted or blocked.

Prepared: D. Early
 Approved: T. Nguyen
 Approval Date: 7/25/00
 Change #: 1
 Directive #: CCBD ME3-01-5638

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Phase	Failure / Effect Description	Criticality Hazard Reference
S 4.1	<p>The preburner gases do not ignite causing fuel pump speed to be below redline values; controller initiates engine shutdown. Mission scrub. Loss of vehicle due to LOX-rich operation may result if failure to establish fuel preburner ignition is not detected.</p> <p>Redundancy Screens: ORIFICE SYSTEM, SENSOR SYSTEM: UNLIKE REDUNDANCY</p> <p>A: Pass - Redundant hardware items are capable of checkout during normal ground turnaround. B: Pass - Loss of a redundant hardware items is detectable during flight. C: Pass - Loss of redundant hardware items could not result from a single credible event.</p>	1R ME-B6S

SSME 1EA/CIL
DESIGN

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Design / Document Reference

FAILURE CAUSE: A: Contamination.

GASEOUS NITROGEN (GN2) PURGE IS USED DURING PROPELLANT CONDITIONING TO REMOVE MOISTURE AND TO DILUTE PROPELLANT LEAKAGE. GASEOUS NITROGEN PURGE PARTICULATES ARE CONTROLLED BY THE INTERFACE CONTROL DOCUMENT (1). CLEANLINESS REQUIREMENTS ARE ESTABLISHED TO REDUCE THE POSSIBILITY OF ORIFICE BLOCKAGE (2). ENGINE SYSTEMS ARE CLEANED TO APPLICABLE MEDIA CLEANLINESS REQUIREMENTS (2). GN2 PURGE IS FILTERED BY THE PNEUMATIC CONTROL ASSEMBLY (PCA). THE GN2 INLET FILTER REMOVES PARTICULATES LARGER THAN 15-MICRON (3). THE PCA DETAIL PARTS AND TEST FIXTURES ARE CLEANED (2) PRIOR TO ASSEMBLY (4). ASSEMBLY AND TEST ARE PERFORMED IN A CLEAN ROOM (5). LUBRICANTS ARE NOT ALLOWED FOR ASSEMBLY OR TEST (4). COMPONENT LEVEL TEST FLUIDS ARE NITROGEN AND HELIUM, WHICH MEET THE HARDWARE CLEANLINESS REQUIREMENTS (2). THE COMPONENT PARTS AND SUBASSEMBLY ARE FREE OF VISIBLE FOREIGN PARTICLES AT THE TIME OF ASSEMBLY (4). THE ORIFICE SIZE IS LARGER THAN ACCEPTABLE PARTICULATES.

THE OXIDIZER SUPPLY IS FILTERED TO 800-MICRONS AT THE EXTERNAL TANK (1). THE ASI SYSTEM HAS BEEN DESIGN VERIFICATION TESTED FOR LOW PRESSURE IGNITION AND LOW MIXTURE RATIOS (6). THE CONTROLLER SOFTWARE IS CONFIGURED TO DETECT AND RESPOND PROPERLY TO FAILURE IDENTIFIED AND COMMAND A SAFE ENGINE STATE (7).

(1) ICD13M15000; (2) RL10001; (3) R0019450; (4) RL00226, RL00347; (5) RQ0711-600; (6) RSS-305-19; (7) CP406R0001 PT 1, 3.2.3.3

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INSPECTION AND TEST

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Failure Causes	Significant Characteristics	Inspection(s) / Test(s)	Document Reference
A	ORIFICE PCA GN2 INLET FILTER		RS009100 RES1090
	CLEANLINESS OF COMPONENTS	THE ASSEMBLY AND UPSTREAM COMPONENTS ARE CLEANED PER SPECIFICATION REQUIREMENTS. AFTER WELDING, THE PASSAGE PORTS AND ORIFICES ARE INSPECTED FOR BLOCKAGE DUE TO WELD MATERIAL. DURING THE PROPELLANT CONDITIONING, THE OXDIZER SYSTEM PURGE IS VERIFIED PER SPECIFICATION REQUIREMENTS. THE SSME PROPELLANT SYSTEM IS DRIED AND VERIFIED DRY PRIOR TO EACH FLIGHT.	RL10001 RL10011 OMRSD S00FB0.300 OMRSD V41CB0.080 OMRSD V41CB0.081
	PCA GN2 FILTER INTEGRITY	FILTERS ARE INSPECTED TO MEET FLOW AND FILTRATION REQUIREMENTS PER SPECIFICATION REQUIREMENTS.	RC1090
	ASSEMBLY INTEGRITY	THE HOT FIRE TESTING AND 2ND E & M INSPECTIONS VERIFY CORRECT OPERATION.	RL00050-04 RL00056-06 RL00056-07
	PRE-FLIGHT CHECKOUT	THE ASI CHAMBERS ARE INSPECTED FOR DAMAGE PRIOR TO EACH LAUNCH.	OMRSD V41BU0.040

Failure History: Comprehensive failure history data is maintained in the Problem Reporting database (PRAMS/PRACA)
 Reference: NASA letter SA21/88/308 and Rocketdyne letter 88RC09761.

Operational Use: Not Applicable.