

NAME P/R QTY	FUNCTION	FAILURE MODE & CAUSES	MISSION PHASE	FAILURE EFFECT	FAILURE DETECTION FLIGHT/GROUND	TIME TO EFFECT/ ACTIONS	CRIT	REMARKS/ HAZARD	REF
DISPLAY AND CONTROLS ELECTRONICS, ITEM 35D 34792291 (1)	Provides current limiting for EVC, feedback solenoid and CLIV solenoid power. Provides optical isolation and discrete signal conditioning for CVS input discretes and EVC tone discretes. Contains battery current and voltage sense circuits, BCU display, and provides secondary power to BCU display, CVS, and sensors.	35849006: Battery power "LOW" discrete fails odd.  CAUSE: Broken connection, open, or electrical component failure.	PRE-EVA  EVA	END ITEM: Lack of indication that battery is providing EMI power.  GPE INTERFACES: False CMB indication of EMI battery configuration. Loss of battery and all consumables monitoring capability by the CMB. During EVA, the CMB fails to enable the low voltage and high battery current limit check, and the consumables management function.  MISSION: None for single failure. Terminates EVA for second failure which prematurely depletes battery, primary oxygen or primary water supply.  CREW/VEHICLE: None for single or double failure. Possible loss of crewman with loss	FLIGHT: Yes. During periodic status checks the following status messages would be bypassed when on battery power: 1. TIME EV: XX 2. XXX 02 (OR PAR) LF TIME LF:XX (limiting consumable).  GROUND: Yes. FEMU-R-001, Para. 7.3.3.2.4.5, EMI Vacuum Performance Matrix, BCU display. Improper message during vacuum chamber run.	None.  TIME AVAILABLE: N/A  TIME REQUIRED: N/A	3/1R  A-PASS B-PASS C-PASS	Redundant paths are the battery power circuits and the SOP. Loss of EVA battery use warning capability.	None.

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 EMU FAILURE MODE, EFFECT ANALYSIS

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ANALYST:

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