

NAME P/N QTY	FUNCTION	FAILURE MODE & CAUSE	MISSION PHASE	FAILURE EFFECT	FAILURE DETECTION FLIGHT/GROUND	TIME TO EFFECT/ ACTIONS	CRIT	REMARKS/ HAZARD	REF
DCM ELECTRONIC ASSEMBLY, ITEM 558 VV92291 (1)	Provides current limiting for EVC, feedwater solenoid and CLIV solenoid power. Provides optical isolation and discrete signal conditioning for DCM input discretes and EVC tone discretes. Contains battery current and voltage sense circuits, DCM display, and provides secondary power to DCM display, ECU, and sensors.	350PH16: Battery current sense circuit output drifts low. CAUSE: Electronic component failure or broken connection.	PREEVA EVA	END ITEM: Displayed battery current lower than actual current. EFE INTERFACE: Battery ampera hour accumulator reads low. Falls to indicate high current draw for subsequent failure. MISSION: None for single failure. False indication that additional EVA time is available. If a subsequent short occurred which increased the current draw, no warning message would be issued. CREW/VEHICLE: None for single (current sensor) or double (high current draw which could prematurely deplete battery) failure. Possible loss of crewman with loss of SOP.	FLIGHT: Yes. Battery failure would cause loss of fan and com. GROUND: Yes. FEPU-2-DD1, Para. 7.3.3.2.1.1. REQUIRED: 10, Transducer and N/A DCM and Gauge Check.	None. TIME AVAILABLE: N/A	I/R A-PASS B-PASS C-PASS	Redundant paths are the battery power circuit and the SCP. The EMS issues a fault message when battery voltage drops below 15.7 VDC.	None.